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I. General

1. Type / Models

Solo 2625/ Solo 2625 01, 2625 02

2. Type Certificate Holder

Solo Kleinmotoren GmbH
Stuttgarter Straße 41
71069 Sindelfingen
Germany

EASA ADOA reference: AP 136

3. Manufacturer

Solo Kleinmotoren GmbH
Stuttgarter Straße 41
71069 Sindelfingen
Germany

4. Certification Application Date

20 September 1996

Note: The application was made to LBA Germany before EASA had been established according to German national procedures.

5. Certification Date

16 March 1998

II. Certification Basis

1. Airworthiness Standards

JAR-22 Change 5 dated October 28. 1995, Subpart H

2. Special Conditions

None

3. Deviations

None

4. Equivalent Safety Finding

None

5. Environmental Protection Requirements:

None (not required for piston engines)

III. Technical Characteristics

1. Type Design Definition

Type Design Definition in accordance with structure parts list 20 00 625 01 of 26 January 1998 (*)

(*) = or later approved revisions

2. Description

Two-stroke two-cylinder inline engine with liquid cooling, contactless dual magneto ignition, pulley for propeller driven the aircraft type and a custom exhaust system.

Displacement : 625 cm³

Bore x stroke : 76mm/69mm

Compression ratio : 9,5:1

Gear ratio: according to the ratio of the engine-propeller-diameter pulleys.

3. Equipment

As stated in the type definition parts list.

4. Dimensions

Model		Solo 2625 01	Solo 2625 02
Overall Length	mm	345	350
Overall Height	mm	408	390
Width	mm	238	255

5. Dry Mass

Model	Solo 2625 01	Solo 2625 02
Mass kg	23 without exhaust system	24 without exhaust system

6. Ratings

Model	Solo 2625 01	Solo 2625 02
Take-off Power	39 KW at 6250 rpm	47 KW at 6500 rpm
Max. Continuous Power	39 KW at 6250 rpm	47 KW at 6500 rpm

Note: The performance value specified above correspond to minimum values defined under the conditions of ICAO or ARDC standard atmosphere.

7. Control System:

The engine model Solo 2625 01 is equipped with one carburettor and a contactless dual magneto ignition. The engine model Solo 2625 02 is equipped with two carburettors and a contactless dual magneto ignition.

8. Fluids:

See Operation and Maintenance Manual for approved fluids.

9. Aircraft Accessory Drives:

None

IV. Operational Limitations

- 1. Temperature Limits**
Cylinder head: Water temperature max. 115 °C

- 2. Maximum/ Minimum Speeds:**

Model	Solo 2625 01	Solo 2625 02
Minimum Cont. Speed	2300 rpm	2300 rpm
Max. Take off and Cont. Speed	6250 rpm	6500 rpm
Maximum Speed	7000 rpm	7000 rpm

- 3. Pressure Limits:**
Fuel supply pressure min. 0,2 bar, max. 0,4 bar.
- 4. Oil Consumption and Capacity Limits:**
See engine manual

V. Operational and Service Instructions

Model	Solo 2625 01	Solo 2625 02
Manual for engine (Handbuch für den Motor)	Solo Typ 2625 01, Issue 1 dated 24.09.1997 (*)	Solo Typ 2625 02, Issue 1 dated 24.09.1997 (*)
Service-Anleitung für den Flugmotor	Solo 2625 01, Ausgabe 1 vom 28.02.1998 (*)	Solo 2625 02, Ausgabe 1 vom 28.02.1998 (*)

(*) = or later approved revisions

VI. Notes

- The manufacturer documents recorded in this TCDS are binding in the specified issue or later approved revisions.
- The suitability and allowable operating ranges of an engine for use in a specific aircraft/propeller combination are to be demonstrated in the aircraft certification.
- For the permitted engine operating hours refer to the relevant operating instructions.
- The deviations of the series 02i with fuel injection are defined in the Service Bulletin No 4600-3, issue 1, dated 30.09.2010.